

شبكة المعلومات الجامعية التوثيق الإلكتروني والميكروفيلو

بسم الله الرحمن الرحيم





HANAA ALY



شبكة المعلومات الجامعية التوثيق الإلكتروني والميكرونيله



شبكة المعلومات الجامعية التوثيق الالكتروني والميكروفيلم



HANAA ALY



شبكة المعلومات الجامعية التوثيق الإلكترونى والميكروفيلم

جامعة عين شمس التوثيق الإلكتروني والميكروفيلم قسم

نقسم بالله العظيم أن المادة التي تم توثيقها وتسجيلها على هذه الأقراص المدمجة قد أعدت دون أية تغيرات



يجب أن

تحفظ هذه الأقراص المدمجة بعيدا عن الغبار



HANAA ALY





AIRFOIL MORPHING CONTROL AT LOW REYNOLDS NUMBER

By

Yusuf Thaer Fadel Yahia Elbadry

A Thesis Submitted to the
Faculty of Engineering at Cairo University
in Partial Fulfillment of the
Requirements for the Degree of
MASTERS OF SCIENCE
in
Aerospace Engineering

AIRFOIL MORPHING CONTROL AT LOW REYNOLDS NUMBER

By

Yusuf Thaer Fadel Yahia Elbadry

A Thesis Submitted to the
Faculty of Engineering at Cairo University
in Partial Fulfillment of the
Requirements for the Degree of
MASTERS OF SCIENCE
in
Aerospace Engineering

Under the Supervision of

Prof. Mohamed Madbouli AbdelRahman Assoc. Prof. Amr Gamal Guaily

Professor

Associate Professor

Aerospace Engineering Department Faculty of Engineering, Cairo University Engineering Mathematics and Physics Department Faculty of Engineering, Cairo University

AIRFOIL MORPHING CONTROL AT LOW REYNOLDS NUMBER

By

Yusuf Thaer Fadel Yahia Elbadry

A Thesis Submitted to the
Faculty of Engineering at Cairo University
in Partial Fulfillment of the
Requirements for the Degree of
MASTERS OF SCIENCE
in
Aerospace Engineering

Approved by the Examining Committee:

Prof. Mohamed Madbouli AbdelRahman,	Thesis Main Advisor
Tion withing was our resultanian,	
Assoc. Prof. Amr Gamal Guaily,	Advisor
Prof. Mohamed Khalil Ibrahim,	Internal Examiner
Prof. Ahmed Farouk Abdelgawad,	External Examiner
(Professor at Zagazig University)	

FACULTY OF ENGINEERING, CAIRO UNIVERSITY GIZA, EGYPT 2022 **Engineer's Name:** Yusuf Thaer Fadel Yahia Elbadry

Date of Birth: 30/1/1996 **Nationality:** Egyptian

E-mail: yelbadry@nu.edu.eg **Phone:** +20 100 570 7861

Address: Opera City, Shiekh Zayed, Giza, Egypt

Registration Date: / 10 /2018 **Awarding Date:** / /2022

Degree: Masters of Science **Department:** Aerospace Engineering

Supervisors:

Prof. Mohamed Madbouli AbdelRahman

Assoc. Prof. Amr Gamal Guaily

Examiners:

Prof. Mohamed Madbouli AbdelRahman

Assoc. Prof. Amr Gamal Guaily Advisor

Prof. Mohamed Khalil Ibrahim Internal Examiner
Prof. Ahmed Farouk Abdelgawad External Examiner

(Professor at Zagazig University)

Title of Thesis:

Airfoil Morphing Control at Low Reynolds Number

Key Words:

Finite Element Method, Incompressible Flow, Morphing, Frequency Lock-In, Level-Set Method

Summary:

Micro Air Vehicles operates at low Reynolds number. The flight can be affected by very small disturbance and the flow can separate easily. New oscillating profile is proposed for oscillating the leading edge of NACA0012 airfoil. The incompressible Navier-Stokes equations are numerically solved using Galerkin/Least-Squares FEM.



Thesis Main Advisor

Disclaimer

I hereby declare that this thesis is my own original work and that no part of it has been submitted for a degree qualification at any other university or institute. I further declare that I have appropriately acknowledged all sources used and have cited them in the references section.

Name:	Yusuf Tha	er Fadel	Yahia	Elbadry	Date:	/	/ 2022

Signature:

Acknowledgements

In the name of Allah the most merciful the most gracious; all thanks to Allah the Lord of the Heavens and Earth and peace be upon prophet Mohamed and his companions.

First of all, I wish to express my deepest gratitude to my main supervisor Professor Mohamed Madbouli AbdelRahman, Professor of Aerodynamics, for all the support, the help he provided, and his guidance during this work. I would like to thank him also for his efforts during the undergraduate studies. It is such an honor to be one of his students. I also wish to express my deepest gratitude to Associate Professor Amr Gamal Guaily, Professor of Engineering Mechanics, for his guidance, for the support, and for giving time for me to teach me. I also wish to express my deepest gratitude to Associate Professor, Mohamed A. Boraey, Professor of Mechanical Engineering, for the support, guidance, and consideration. Dr. Mohamed has offered his ongoing help on both the personal issues and the technical problems that I have faced. Dr. Amr and Dr. Mohamed have been more than elder brothers to me, and literally, words can not describe their efforts.

Words can not describe how I want to thank my family for all what they provided for me. I would not have become the one who I am without their support. I have always found love and motivation from my father, my mother, and my sisters.

In the end, I must express my thanks to my brothers Mohamed Adawy, Mohamed Gamal, and Ahmed Hamada for all the support during the past years. I also wish to thank my friends in Aerospace Department, Mina Naem and Karim Mohamed, Hossam Hassan, Mahmoud Ayyad, and Ahmed Bayram for all the guidance and advice they provided during this work. Also, I must thank my friends in Nile University, Omar Khaled, Moustafa AbdelHamid, Mohamed Hesham, Mohamed Abdallah, Haytham Yassin, Menatallah Soliman, Mostafa Abdelrahman, Mahmood Saleh, Ahmed Saeed, Hossam Hassan, Lamia Zain, Mohamed Diab, Ahmed Ashraf, and Bahaa for all the support during the past three years.

Yusuf Elbadry

Table of Contents

Di	sclain	ner	i
Ac	know	ledgements	ii
Li	st of T	Tables	v
Li	st of I	Figures	vi
Li	st of A	Algorithms	X
No	omeno	elature	xi
Al	ostrac	t	xiii
1	Intro	oduction and Literature Review	1
	1.1	Low Reynolds Number Aerodynamics	1
		1.1.1 Introduction to micro air vehicles	1
		1.1.2 Airfoil Characteristics at Low Reynolds Numbers	2
		1.1.3 Passive and active control methods	4
		1.1.3.1 Passive flow control methods	4
		1.1.3.2 Active flow control methods	5
	1.2	Fixed grids and capturing methods	7
	1.3	Level-Set Method	7
	1.4	Objectives of the Thesis	9
2	Nun	nerical Solver of Navier-Stokes Equations	10
	2.1	Unsteady incompressible Navier—Stokes equations	10
		2.1.1 Continuity equation	10
		2.1.2 Momentum equations	10
	2.2	Finite Element formulation	11
		2.2.1 Spatial discretization	
		2.2.2 Matrix formulation	
	2.3	Solver Algorithms	15
3		dation	19
	3.1	Solver Information	19
	3.2	Two-dimensional unsteady incompressible	
		Navier–Stokes equations	19
		3.2.1 Governing equations	19
		3.2.2 Lid-driven cavity problem	20
		3.2.2.1 computational domain and boundary conditions	20
		3.2.2.2 results and discussion	21
		3.2.3 Flow around a two-dimensional cylinder	25
		3.2.3.1 Computational domain and boundary conditions	25

			3.2.3.2 Results						
		3.2.4	Flow over oscillating cylinder	27					
			3.2.4.1 Computational domain and boundary conditions	28					
			3.2.4.2 Results	29					
		3.2.5	Flow over stationary airfoil	32					
			3.2.5.1 Computational domain and boundary conditions	32					
			3.2.5.2 Results	33					
4	Peri	Periodic Morphing Control							
	4.1	Morph	hing Technique	34					
	4.2	Valida	ation of Lock-In Frequency	35					
		4.2.1	Mesh independence test	35					
		4.2.2	Time step and solver parameters	35					
		4.2.3	Grid generation	36					
		4.2.4	Results of frequency lock-in	37					
	4.3	Effect	of Changing the Reynolds Number and the Angle of Attack on the						
		freque	ency Lock-in Region	44					
		4.3.1	Case 1: Re=6000 and $\alpha = 6^{\circ}$	44					
		4.3.2	Case 2: Re=4000 and $\alpha = 6^{\circ}$	46					
		4.3.3	Case 3: Re=5000 and $\alpha = 7^{\circ}$	48					
		4.3.4	Case 4: Re=5000 and $\alpha = 4^{\circ}$	51					
	4.4	Effect	of the oscillating position	55					
	4.5	Propos	sed new morphing profile	58					
5	Sum	mary a	and Conclusion	62					
	5.1	Thesis	s Summary	62					
	5.2	conclu	usion	63					
	5.3	Future	e Work	63					
		5.3.1	Three Dimensional Model	63					
		5.3.2	Turbulence Model	64					
		5.3.3	Arbitrary Lagrangian Eulerian Algorithm	64					
		5.3.4	Modification to the solver	64					
Re	eferen	ces		65					

List of Tables

4.1	Effect of Mesh variation on mean and maximum C_l	35
4.2	Effect of time step variation on mean and maximum C_l	36

List of Figures

3.22 3.23	Lift and drag coefficients for different excitation frequencies - current work Grid around the airfoil	31 32 32 33
4.1	Kang et al. [34] deformation profile for the leading edge morphing. Where the x-axis represents the non-dimensional x-coordinate $x=(\tilde{x}/c)$ and the y-axis represents the non-dimensional y-coordinate $y=(\tilde{y}/c)$	34
4.2	Grid around leading edge of NACA 0012 Airfoil	36
4.3	Velocity and vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of attack of 6°	36
4.4	Variation of the primary and the secondary frequency with the excitation frequency	37
4.5	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of	υ,
	attack of 6° at excitation frequency $f_e = 0.41.$	38
4.6	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of	
4.7	attack of 6° at excitation frequency $f_e = 0.6$ Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of	38
4.0	attack of 6° at excitation frequency $f_e = 0.8.$	39
4.8	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of	39
4.9	attack of 6° at excitation frequency $f_e = 1.0.$	39
	attack of 6° at excitation frequency $f_e = 1.2$	39
4.10	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of	
	attack of 6° at excitation frequency $f_e = 1.4$	40
4.11	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of	4.0
4.12	attack of 6° at excitation frequency $f_e = 1.667$	40
4.12	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of attack of 6° at excitation frequency $f_e = 1.76$	40
4 13	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of	70
1.13	attack of 6° at excitation frequency $f_e = 1.9.$	41
4.14	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of	
	attack of 6° at excitation frequency $f_e = 2.1.$	41
4.15	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of	
	attack of 6° at excitation frequency $f_e = 2.3.$	41
4.16	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of	40
1 17	attack of 6° at excitation frequency $f_e = 2.5.$	42
4.1/	Vorticity contours around NACA0012 airfoil at $Re = 5000$ and angle of attack of 6° at excitation frequency $f_e = 2.71.$	42
4 18	Spectral analysis of flow with different oscillating frequencies at $Re = 5000$	72
10	and angle of attack 6° at excitation frequencies (a) 0.6, (b) 1.2, (c) 1.667,	
	(d) 2.71	43
4.19	Comparison between the excitation frequency and the primary and sec-	
	ondary frequencies of the flow with $A_o = 0.00365$ and $A_e = 0.00222$ at	
	Re=6000 and $\alpha = 6^{\circ}$	44

4.20	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 6000$	
	and angle of attack of 6° at $f_e = 0.4$ with $A_o = 0.00365$ and $A_e = 0.00222$.	45
4.21	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 6000$	
	and angle of attack of 6° at $f_e = 0.6$ with $A_o = 0.00365$ and $A_e = 0.00222$.	45
4.22	Vorticity Contours and streamlines around NACA0012 airfoil at $Re = 6000$	
	and angle of attack of 6° at $f_e = 1.4$ with $A_o = 0.00365$ and $A_e = 0.00222$.	45
4.23	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 6000$	
	and angle of attack of 6° at $f_e = 1.8$ with $A_o = 0.00365$ and $A_e = 0.00222$.	46
4.24	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 6000$	
	and angle of attack of 6° at $f_e = 2.5$ with $A_o = 0.00365$ and $A_e = 0.00222$.	46
4.25	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 6000$	
	and angle of attack of 6° at $f_e = 2.7$ with $A_o = 0.00365$ and $A_e = 0.00222$.	46
4.26	comparison between the excitation frequency and the primary and sec-	
	ondary frequencies of the flow with $A_o = 0.00365$ and $A_e = 0.00222$ at	
	Re=4000 and $\alpha = 6^{\circ}$	47
4.27	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 4000$	
. • •	and angle of attack of 6° at $f_e = 0.4$ with $A_o = 0.00365$ and $A_e = 0.00222$.	47
4.28	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 4000$	40
4.20	and angle of attack of 6° at $f_e = 1.5$ with $A_o = 0.00365$ and $A_e = 0.00222$.	48
4.29	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 4000$	40
4.20	and angle of attack of 6° at $f_e = 1.7$ with $A_o = 0.00365$ and $A_e = 0.00222$.	48
4.30	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 4000$	40
4 2 1	and angle of attack of 6° at $f_e = 2.7$ with $A_o = 0.00365$ and $A_e = 0.00222$.	48
4.31	comparison between the excitation frequency and the primary and sec-	
	ondary frequencies of the flow with $A_o = 0.00365$ and $A_e = 0.00222$ at	49
1 22	Re=5000 and $\alpha = 7^{\circ}$	49
4.32	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 5000$	49
1 22	and angle of attack of 7° at $f_e = 0.4$ with $A_o = 0.00365$ and $A_e = 0.00222$. Vorticity contours and streamlines around NACA0012 airfoil at $Re = 5000$	49
4.33	and angle of attack of 7° at $f_e = 0.6$ with $A_o = 0.00365$ and $A_e = 0.00222$.	50
1 31	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 5000$	50
7.54	and angle of attack of 7° at $f_e = 1.4$ with $A_o = 0.00365$ and $A_e = 0.00222$.	50
4 35	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 5.00222$.	50
т.ЭЭ	and angle of attack of 7° at $f_e = 1.8$ with $A_o = 0.00365$ and $A_e = 0.00222$.	50
4 36	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 5.00222$.	50
1.50	and angle of attack of 7° at $f_e = 2.5$ with $A_o = 0.00365$ and $A_e = 0.00222$.	51
4.37	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 5000$	<i>J</i> 1
	and angle of attack of 7° at $f_e = 2.7$ with $A_o = 0.00365$ and $A_e = 0.00222$.	51
4.38	comparison between the excitation frequency and the primary and sec-	-
	ondary frequencies of the flow with $A_o = 0.00365$ and $A_e = 0.00222$ at	
	Re=5000 and $\alpha = 4^{\circ}$	52
4.39	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 5000$	
	and angle of attack of 4° at $f_e = 0.4$ with $A_o = 0.00365$ and $A_e = 0.00222$.	52
4.40	Vorticity contours and streamlines around NACA0012 airfoil at $Re = 5000$	
	and angle of attack of 4° at $f_{e} = 0.5$ with $A_{o} = 0.00365$ and $A_{e} = 0.00222$.	53